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JUNE 14, 1926

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VOLUME
XX

SPECIAL FEATURES

NUMBER
24

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MORE ARCTIC FLIGHT PHOTOGRAPHS
CORRUGATED METAL SHEET IN AIRCRAFT CONSTRUCTION

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AVIATION

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Selling the Air Mail

THE SUCCESS or failure of the privately sponsored air mail lines which are starting up throughout the country depends primarily on the question of whether they get a sufficient amount of revenue to meet normal operating expenses. Regularity and economy is extremely important and depends on the management of the lines almost as much as it does on the equipment used but, even if a line were perfectly managed and had the best of equipment and pilots, it could not survive if it did not have the revenue. Where the management or equipment is bad but the line shows that it is getting enough revenue to meet the normal expenses which are reflected by the operations of other lines, it is always possible to get new management, better equipment and sufficient working capital. The question of revenue is paramount but because the operations of the planes is more interesting and romantic it has absorbed the attention of the organizers of the new air lines and they have not concentrated on the main problem—getting the revenue.

It is hard for the untrained individual, who absorbs with eager interest every bit of aeronautical news, to realize how important and indifferent the general public is to the existence of the air mail. The average business man or, more important still, his congressman, has a hazy idea that there is an air mail service but rarely knows anything definite about the matter and does not realize how vital the air mail might be to his particular requirements, and, therefore, does not bother to find out what it is all about. The value of the air mail must be drummed into the American public through a steady campaign of publicity and advertising. Air mail is a commodity like any other and it must be sold to the business man and, therefore, constantly kept before his attention.

If the air lines are to succeed they must realize that the sales cost is a part of their operating expense and is just as vital to the existence of their organizations as is the gasoline which goes into the engines of their planes. They must further realize that the effort must be national. For example, the Salt Lake-Los Angeles line will get much of its business from New York and vice versa. Normal sales methods of free publicity, advertising and circulation can be used but they all cost money and each air line must bear its proportional share of expense. Air lines can do a lot through local Chambers of Commerce and other civic and business organizations but the great task of selling the air mail to the American public can only be done through a great and continued nationwide effort.

Such an effort can only be accomplished through the cooperation of all the privately owned air lines. Each line striving individually will work up a certain amount of business for itself locally, but as much as its business must come from far away points it can be entirely successful only by advertising the whole country to the value of the air mail and the ways to use it.

Last fall some of the private operators got together and held a meeting for the purpose of outlining methods of getting business. Many of the future mail lines were not represented and very little was accomplished but by next fall the operators will have begun to realize the necessity of closer cooperation and a meeting would probably accomplish certain very definite results. In the meantime it is up to every one interested in aviation to follow the lead of the Chamber of Commerce, the N. A. A. and many of the flying clubs and to boost the use of the air mail in every way possible.

Airplane and Automobile Cooperation

THIS CLOSELY allied connection which exists between certain sections of engineering art is perhaps in no other case so clearly apparent as it is between the automobile and the aeronautical fields, particularly in respect to the airplane. While there are certain specific requirements of paramount importance in each and every field, the very considerable overlapping, which must of necessity exist in aeronautical and automobile engineering has undoubtedly been very largely responsible for the comparatively rapid and, even, at times, phenomenal development that has taken place during the past twenty years in aeronautics.

There was a time, in the early days of flying, when the debt which the airplane held to the automobile was extremely great, whereas the gasoline engine, which was primarily produced for road transportation, the airplane would have been impossible. That aeronautics is, however, simply paying back this debt is clear when the remarkable developments that have been made in airplane engines are considered—developments which cannot but have a very vital effect upon the reliability, economy and general success of the automobile.

That an even closer cooperation may now exist between automobile and aeronautical engineers was a thought indirectly expressed by J. T. Liffie, Jr., president of the Society of Automobile Engineers, at the Spencer meeting of this body held at French Life Springs, Ind., during the week of May 31. In the opinion of Mr. Liffie, automobiles must become smaller and lighter. The problems of the aeronautical engineer have ever been mainly concerned with the question of weight and it would seem highly probable that much of the development work which has been carried out in this direction, not only in the case of engine design but also concerning the design of light rigid structures, capable of withstanding heavy loads, will prove of value to automobile designers.

This is in reality an extremely interesting outlook because, with a great host of automobile engineers bending their energies in the direction of obtaining lighter metal structures, involving the possible extensive use of aluminum and titanium in automobile construction, the designers stand but benefit from the added concentration on a common subject of such importance.

structured tubular steel framework which could be quickly fixed just outside the gondola.

A special type of ballast was fitted in the base of the gondola for the landing aerial, which was 300 ft. long. This ballast enabled a new aerial and weight to be fitted while the airplane was in flight should it be necessary.

Receiver

Special Marconi receiving apparatus was utilized both for distances flying and for emergency service reception. A short five-two-receiver unit with a wave range of 18 to 180 meters was carried. The aerial for this apparatus was a short length of wire fixed between the wireless cabin and one of the engine gondolas.

The direction finder loops were fitted diagonally round the outside of the gondola, the center of the loops coinciding with the fuselage in the radio cabin. The loops consisted of two turns of wire, spaced about two inches apart. The loops were doped to the fabric with thin tape, between a seal and sealant which was very efficient fitting.

The amplifier was provided with six type T 2 valves with anode-cathode transformer coupling for high frequency magnification and a type QX valve for rectification.

For the reception of continuous wave, spark and telephone screen messages a "plug-in" unit carrying a wave range of 300 to 25,000 meters was connected to the high frequency amplifier. The tuner was a coupled resonant instrument provided with constant coupling to the grid circuit.

Commercial Aircraft Conference

On May 26 a joint conference of the Sub-Committee on Aerodynamics of the National Advisory Committee for Aeronautics with representatives of aircraft manufacturers and operators was held at Langley Field. The meeting was the first of this nature held since 1915. Some of the party were direct in Langley Field, but the majority went by boat to Old Point Comfort and arriving there at 9:45 a. m., and leaving for the Field at 8:25 a. m., after breakfasting at Starwood Inn. A preliminary meeting was held in the Officers' Club after the party reached its destination, followed by an inspection of the atmosphere wind tunnel, variable density tunnel, power plants laboratories, engine houses, propeller research equipment, and instruments. After lunch which was served in the N.A.C.A. dining room, a joint conference of the Sub-Committee and the representatives was held, during which several new projects were discussed and a review made of commercial aviation activities. The party left for Washington after a demonstration of Ames research and bombing was given and a tour of the Field was made.

Among those present were: Charles L. Lawrence, president of the Wright Aircraft Corporation; J. T. Hoot, vice of the Vought company; W. Mitchell, manager of the Ford plant in Norfolk; J. P. Farney, assistant general secretary of the N. A. C. A., Washington; James G. Hay, operations

The chief difficulty in fixing the apparatus in the radio cabin of the Kings was lack of space, the operator's quarters being entirely somewhat cramped. In fixing the instruments it was necessary to give more attention to convenience of wiring than to appearance, but it is quite of every necessity in space and weight the greatest effort was very workable. The receiving apparatus was fixed on two shelves as shown the center on the interior wall of the cabin. The whole forward wall of the cabin was taken up with the transmission panel.

A narrow table was provided for the operator's writing and for the accompanying log on the stationary outside wall. Under the left hand side of the table the transmitter battery was fitted to the floor. A vacuum for the gas transmitting were adjusted was also fixed on the left under the table. Next to these instruments, on the right hand side, was fitted a small rectangular cupboard for carrying spares.

The transmitters and dry batteries stood on the floor. A double plate change over switch was fitted in the center and the 15-volt lighting system for the ship was brought in this as well as the radio 12-volt supply. This enabled the ship's accumulator and the radio accumulator to be changed in parallel from the radio generator, and if the ship's was gone out, current could be supplied from the radio accumulator and vice versa. The accumulator used for the ship's motor and for the Marconi apparatus was of the thin plate high discharge type and not updatable.

members of the Florida Aviation Co.; Karl Amstein, vice-president of the Goodwin-Baggett Corp.; Hugh L. Boyden, of the Bureau of Standards; William H. Short, of the Aviation Section, Ford Motor Co.; W. A. Hinkoff, of the Aircraft Development Corp.; W. G. Brunsbacher, of the Bureau of Standards; A. E. Sells, of the Panhandle Aircraft Corp.; J. P. Wright, of the Curtis Aeroplane & Motor Co., Inc.; G. F. Pope, secretary of Ohio Wood Hall, Inc.; Harold F. Parsons, Pitman Aviation Co.; Arthur E. Larson and Robert W. A. Brown, of the same concern; R. E. Johnson, McCord Field; C. C. Thompson, Jr., of the Bureau of Standards; Gaudy E. E. Wilson, T. S. S. Bureau of Aeronautics; Hugh O. Wadsworth, U. S. A. B. commanding officer, Langley Field; and Dr. W. G. Brunsbacher, Bureau of Standards.

The membership of the aerodynamics committee is as follows: Dr. Joseph S. Ames, Johns Hopkins University, chairman; Gaudy E. C. Richardson, U. S. N., vice-chairman; Dr. E. J. Brugger, Bureau of Standards; Lewis E. W. Bohannon, Jr., and W. S. Dool, U. S. N.; Dr. G. W. Lewis, Bureau of Aeronautics; Roscoe W. Johnson, National Advisory Committee for Aeronautics; Major Leslie McDill, A. B.; Prof. Charles F. Munn, Chief of U. S. Weather Bureau; Dr. A. P. Sols, Construction Dept., Washington Navy Yard; and Prof. Edward P. Warner, Mass. Inst. of Tech.



Members of the National Advisory Committee for Aeronautics and representatives of the aircraft industry at the conference recently held at Langley Field.

Corrugated Metal Sheet in Aircraft Structures

Possibilities of Employing Corrugated Duralumin for Large Aircraft Components in Addition to Parts of Compound Structures.

THE USE of duralumin in aircraft construction is becoming practice with many airplanes designers while the development of this alloy of aluminum has been very largely impossible for the moment in the construction of large metal airships. While there are undoubtedly disadvantages in the employment of duralumin for many classes of aircraft construction, an very favorable Strength/Weight ratio enables this material presently adaptable in the constant field. Duralumin has been employed largely in sheet form in airplane construction and it will be realized that the proposed airship to be produced by the Aircraft Development Corp. will be of duralumin construction throughout, the entire skin being composed of duralumin sheet. Other

It is pointed out that corrugated sheet has been employed in building construction for many years. As an example,



Plate M. A corrugated metal sheet being rolled under a rolling mill.

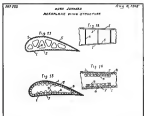


Plate J. Early Junkers metal wing patch.

forms in which duralumin has entered largely into aircraft structures are the numerous strip sections and also in the form of both round and square tube. A type which has been employed as a stressed element and which offers wide possibilities for further development is that of corrugated sheet material. The stage over which corrugated duralumin or other aluminum alloy may be employed in aircraft manufacture has been studied by F. E. Lowy, an experienced engineer of Cleveland, Ohio, and he has prepared a series of notes and a number of interesting photographic collages to the various possibilities of this class of material in aircraft structures.

a patch was used as long ago as 1911 for constructing fire-proof buildings of multi-story of corrugated iron with their corrugations arranged in right angles and welded at the joints of corner. However, the employment of corrugated metal construction in airplane skins took but a few years. In 1903, the German designer, Herr Hugo Junkers advanced the application of corrugated sheet metal to the construction of all-metal airplane wings. Reports from the Junkers patent show, in Plate I, figs. 11 and 12, the method of combining a flat outer skin with a corrugated reinforcement sheet and metal ribs to maintain the section, while in figs. 13 and 14, a ribbon type of metal wing construction is shown, which involves the employment of metal stiffening. Junkers patents were the first to employ corrugated metal construction in both wings and as a fuselage structure, in such use the latest stiffening consisting completely of corrugations in the extent of the internal structure. This class of construction has later been adopted in this country as the sheet all metal monoplanes.

Mr. Lowy notes that some years ago, Stuart Hughes, one of the leading constructors of aircraft in England adopted corrugated wing wing sections in one of his planes. The compound span consisted of an upper and lower beam of



Plate III. Two forms of compound ribs with corrugated metal walls for the control surface of the new wing model 251.

with "U" section metal with a corrugated metal web. The spar was employed such that left flanges were taken at right angles to the plane of the corrugations. A single spar of this construction was directly tested on a loaded beam suspended at each extremity. An accompanying photograph (Plate IX) shows the type of failure which resulted when the



Plate IV Method of applying corrugated metal to wing ribs.

corrugated load, applied at 27 in. from one end, resulted 835 lb. The spar was 120 in. long, 6 in. deep with an aluminum alloy web of corrugated sheet .015 in. thick.

In Airship Structures

This type of construction is admirably suited to the construction of airship spars and booms and has, in fact, been so employed in the construction of the control car of the U.S. Army rigidly inflated dirigible, R-51. Examples of these structural parts of the control car are shown in Plate III which gives a very good idea of the structural details of two forces of compound ribs of the R-51 control car.



Plate V A corrugated hydrostatic floor under static test. The floor sustained a static load of 250 lb. per sq. ft.

In airship construction, the employment of corrugated metal in the structure is fairly common, especially in the construction of built up metal girders of the new self-buoyant type. This particular type of structure is advocated by Mr. Leady as adaptable in the building of airplane wing ribs where all webs are wanted and the rib expansion controlled by means of an irregular Warren truss system derived of metal lattices with longitudinal corrugations, as depicted in Plate IV.

Applicable to Large Structures

That the adaptation of the corrugated metal sheet system is very well suited to complete structures and is not necessarily confined to small units in compound structures, is indicated

by the fact that large surfaces, such as floors for shipboard use, may be made to carry very large loads. Plate V shows a static test upon a specimen of flooring of this type measuring 36 in. by 36 in. and composed of an upper sheet of duralumin of .015 in. thickness and with $\frac{1}{2}$ in. corrugations and a lower steel .048 in. thick with $\frac{3}{8}$ in. corrugations. The flooring sustained a static load of 250 lb. per sq. ft. without failure and itself weighed only 622 lb. per sq. ft. Furthermore, in Plate VI a duralumin floor roadway

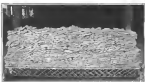


Plate VI Duralumin floor roadway with corrugated metal surface under static test of 300 lb. per sq. ft.

girders with a corrugated duralumin surface is shown which failed under a static load of 500 lb. per sq. ft. The girder was 120 in. long, of equilateral triangular section of 21 in. side. The broad web of .315 in. sheet with $\frac{1}{4}$ in. corrugations, while the webs were .018 in. gauge plates. The complete girder weighed 13 lb. per sq. ft. side.

Calculating Loads

Mr. Leady advocates the following equation for calculating the ultimate load which may be carried by corrugated sheet or lathing, and appends results of tests on corrugated aluminum alloy sheets, as follows:

$W =$ Total ultimate ultimate load, in pounds
 $L =$ Corrugation length of sheet, in inches
 $t =$ Thickness of sheet, in inches
 $W = \frac{L \times t \times 1000}{1000}$

$W =$ Total ultimate ultimate load, in pounds
 $L =$ Corrugation length of sheet, in inches
 $t =$ Thickness of sheet, in inches
 $W = \frac{L \times t \times 1000}{1000}$

RESULTS OF TEST ON CORRUGATED SHEET

Specimen	Length	Depth	Area	Ultimate Load	Ultimate Stress	Yield Stress
1	120 in.	1/2 in.	144 sq. in.	12,500 lb.	87,500 psi	60,000 psi
2	120 in.	1/2 in.	144 sq. in.	11,500 lb.	80,555 psi	55,000 psi
3	120 in.	1/2 in.	144 sq. in.	11,000 lb.	76,388 psi	52,000 psi
4	120 in.	1/2 in.	144 sq. in.	10,500 lb.	72,916 psi	50,000 psi
5	120 in.	1/2 in.	144 sq. in.	10,000 lb.	69,444 psi	48,000 psi
6	120 in.	1/2 in.	144 sq. in.	9,500 lb.	66,027 psi	46,000 psi
7	120 in.	1/2 in.	144 sq. in.	9,000 lb.	62,611 psi	44,000 psi
8	120 in.	1/2 in.	144 sq. in.	8,500 lb.	59,194 psi	42,000 psi
9	120 in.	1/2 in.	144 sq. in.	8,000 lb.	55,777 psi	40,000 psi
10	120 in.	1/2 in.	144 sq. in.	7,500 lb.	52,361 psi	38,000 psi

SINGLE CORRUGATED STRIPS

Specimen	Area	Length	Depth	Area	Ultimate Load	Ultimate Stress	Yield Stress
1	144 sq. in.	120 in.	1/2 in.	144 sq. in.	12,500 lb.	87,500 psi	60,000 psi
2	144 sq. in.	120 in.	1/2 in.	144 sq. in.	11,500 lb.	80,555 psi	55,000 psi
3	144 sq. in.	120 in.	1/2 in.	144 sq. in.	11,000 lb.	76,388 psi	52,000 psi
4	144 sq. in.	120 in.	1/2 in.	144 sq. in.	10,500 lb.	72,916 psi	50,000 psi
5	144 sq. in.	120 in.	1/2 in.	144 sq. in.	10,000 lb.	69,444 psi	48,000 psi
6	144 sq. in.	120 in.	1/2 in.	144 sq. in.	9,500 lb.	66,027 psi	46,000 psi
7	144 sq. in.	120 in.	1/2 in.	144 sq. in.	9,000 lb.	62,611 psi	44,000 psi
8	144 sq. in.	120 in.	1/2 in.	144 sq. in.	8,500 lb.	59,194 psi	42,000 psi
9	144 sq. in.	120 in.	1/2 in.	144 sq. in.	8,000 lb.	55,777 psi	40,000 psi
10	144 sq. in.	120 in.	1/2 in.	144 sq. in.	7,500 lb.	52,361 psi	38,000 psi

DOUBLE CORRUGATED RESULTS

Specimen	Area	Length	Depth	Area	Ultimate Load	Ultimate Stress	Yield Stress
1	144 sq. in.	120 in.	1/2 in.	144 sq. in.	12,500 lb.	87,500 psi	60,000 psi
2	144 sq. in.	120 in.	1/2 in.	144 sq. in.	11,500 lb.	80,555 psi	55,000 psi
3	144 sq. in.	120 in.	1/2 in.	144 sq. in.	11,000 lb.	76,388 psi	52,000 psi
4	144 sq. in.	120 in.	1/2 in.	144 sq. in.	10,500 lb.	72,916 psi	50,000 psi
5	144 sq. in.	120 in.	1/2 in.	144 sq. in.	10,000 lb.	69,444 psi	48,000 psi
6	144 sq. in.	120 in.	1/2 in.	144 sq. in.	9,500 lb.	66,027 psi	46,000 psi
7	144 sq. in.	120 in.	1/2 in.	144 sq. in.	9,000 lb.	62,611 psi	44,000 psi
8	144 sq. in.	120 in.	1/2 in.	144 sq. in.	8,500 lb.	59,194 psi	42,000 psi
9	144 sq. in.	120 in.	1/2 in.	144 sq. in.	8,000 lb.	55,777 psi	40,000 psi
10	144 sq. in.	120 in.	1/2 in.	144 sq. in.	7,500 lb.	52,361 psi	38,000 psi

The Vickers-Wibault Fighter

An All-Metal Single-Seater Pursuit Plane of French and British Origin.

A N INTERESTING new single seater fighter airplane is now being produced by the well-known British firm of Vickers Ltd. It is understood that a number of dummy machines are being built for the Indian Government. The plane is known as Type 721 and was designed by M. Michel Wibault. It is a high-wing biplane monoplane of all-metal—duralumin—construction, in which the requirements of simplicity of structure, maintenance, and repair have been given paramount consideration. From a close inspection of the photographs, it would appear that the propeller is not suited to the normal wood type.

The wing is of thick section and is of purely rectangular shape with the exception of that part on only over the pilot's cockpit. The section, however, appears apparently to be fusiform toward both the wing tips and the center line from the point of the leading and trailing edges. This variation in thickness is so designed as to give a super depth varying such that the strength of the beam is uniformly distributed according to the loading desired.



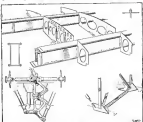
A portion of the Wibault wing showing the rib and leading edge.

The simplicity of the wing structure is notable. The main spars are of box construction. The side members are flanged top and bottom and the top and bottom members are plain metal strips riveted to the flanges. These members are double throughout. The ribs, which are composed of three individual and separate spars, are secured to the spars by means of vertical flanges riveted to the ribs at intervals. The ribs project above the normal profile and corrugate sheets, which have flanged-up ribs are riveted through these edges and the ribs, thus both ends of the ribs are exposed. Furthermore, the ribs do not extend to the leading edge, this being formed of a slanted strip of duralumin riveted to the covering of the rest of the wing.

The ribs themselves consist of simple sheets of metal laminated and strengthened where necessary. Being made in three pieces, they are secured to the spars in the following manner: First, the forward section, riveted to a flange on the

side of the front spar, then the middle section riveted to the corresponding flange the other side of the spar. This middle section is secured in a flange on the rear spar in a similar way and finally the rear section at the rib secured to the rear of the rear spar on a flange. Being made in the wing structure are of duralumin tube carried from stiff brackets riveted to the main spars.

The wing is held in two sections joined on the center line. The rear spar joint is a simple hinge but the front spar joint consists of hinge-like flanges on the top and bottom of the spars joined together with two heavy bolts. The entire structure



Section of the structure.

General structural details of the Wibault. At the top, the general outline of wing structure is shown with on the left a section of the spar, and on the right corner, the method of flange the covering to the ribs. Below left, the arrangement of the central lower showing the spar and push ribs for the covering. Right, fuselage details of the engine and T section structure.

has been laminated in an every rigid in tension and entirely satisfactory.

Unlike the wings, which are, to a large extent, dependent upon the covering for rigidity and strength, the fuselage, though metal covered, has a robust metal frame structure.



The Vickers-Wibault single seater fighter (Daimler engine).

The undercarriage is of the skid-landing type and is of steel construction. Shock absorbers are provided by the Vickers air-cushion shock absorbers. The struts for the undercarriage at the fuselage are above the attachment, for the wing housing struts which extend from the fuselage across the wing to points about two-thirds of the wingspan from the center line on each side. These housing struts consist of parallel streamer tubes. The center housing and support for the wing struts is formed of a ribbed of streamer tubes, as seen in the drawing.

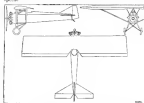


The tailplane is of triangular plan form and has but one wing, which is parallel to the leading edge. The elevators are built on a wing tube kept to the main structure. This tube serves the purpose of the rear spar of the tailplane. The tailplane is braced by two tubes, one on each side, on the outer side. These tubes support the structure of the tail plane front spar and the hinge tube side where their outside at the tip of the tailplane. The tailplane is adjustable in flight by raising or lowering the front spar around the elevator hinge tube at center.

The ailerons are of very narrow chord but extend over the entire span being driven at the middle where the wing is not away over the plan's surface. They are operated entirely from the rear end of the fuselage from one place which the ailerons are built. The control stick operates the ailerons through push and pull indicator rods operated through stepped gears. Elevator and aileron controls are connected to turn on a key shaft in the end of the fuselage through cables. From one of these cables one push-and-pull rod runs to a control elevator lever in the fuselage. From the other a

similar rod runs to a lever on the starboard side of the rudder. The engine is a 120 hp air-cooled radial Jupiter mounted upon a metal bulk plate which is detachable with the engine. Gasoline and oil tanks are of duralumin, entirely from firm welded joints. The joints in the tanks are made between separate flanges of individual sheets. One of each pair of flanges is made twice as deep as the other and then bent around the first flange and riveted. The joints are sealed with a hot-poured sealant.

In performance the Vimy is reported to have functioned well, and the plane is planned to fly. The take-off is quick and the climb and speed at altitude good. It is also said that, in spite of the wing loading (twenty 14 lb. per sq ft.), the machine can be flown under perfect control at very low speeds.



General outline drawings of the Vickers Vimy

The general characteristics of the plane are as follows:

Type	52 ft. 6 in.
Length	29 ft. 6 in.
Wing span	52 ft. 6 in.
Wing area	1,172 sq. ft.
Weight empty	11,410 lbs.
Weight loaded	12,810 lbs.
Power loading	12.5 hp 17 mi.
Speed at 14,000 ft.	120 mph
Climb	12,000 ft.
Endurance	15 hr.



The Vickers Vimy at altitudinal angle under flight; recently performed in England, climbed rapidly at 420 ft

The machine is fitted with a Bristol Jupiter radial en-

New York-Buenos Aires Flight

The Buenos Aires, the Buenos airplane which Berardo Duggan and two Indian pilots are using in their flight from New York to the capital of the Argentine Republic, during the first four days of the journey encountered conditions of storms and winds. From Midway Field the first stage of the flight led to Hampton Beach, successful except for some minor engine trouble and rain. Strong winds caused the plane to spend the night at Charleston, S. C., instead of continuing to Florida, as had been planned for the second stage.

The third part of the flight, from the U. S. Naval Station at Charleston, to Miami, was covered in 4 hr. 12 min., being the longest of the three legs. The maximum of cruising speed was maintained throughout this jump, in spite of the fact that the storms continued. The start of this leg was unusually favorable from the weather standpoint. The run was smoother and wind, very fair. The storm was sudden and intense. With main sailing and the wind directly on the plane's face, the plane flew into clouds of some density. Up to this time the plane had been cruising at an altitude of 2,000 ft. and after construction it was decided to take the Buenos Aires to an altitude of 2,000 ft., which brought it out of the clouds. At Jacksonville the plane descended through the clouds.

The fourth day of the flight, from the Florida Keys into Havana, began at 7:30 a.m. on a heavy fog. Directly over the Keys, Captain Oliver detected a fast-approaching mass of clouds, and in accompanying headwind brought a decision to return to Keyport Bay. A lower level was made and at a speed of 125 m.p.h. with the storm wind behind, the plane sped to its harbor. The flight to the Keys and return to Keyport Bay took 1 hr. 45 min. Weather, continued unfavorable, and although the plane had not suffered from its experience, it was decided to wait until the morning for the trip to Havana.

The favorable weather of May 25 gave the team the opportunity for the delayed flight from Keyport Bay to Havana. Under the most pleasant conditions encountered up to this time, the plane made this leg of the trip in 2 hr. and 35 min.; the members of the party receiving an enthusiastic welcome in the Cuban capital.

Havana to Port au Prince, and from that city to San Juan has been a steady record of storms and winds. The only important happening was a collision with a launch in the harbor at Port au Prince, causing a delay of a day. The U. S. Marine Squadron made necessary repairs, permitting the team to reach San Juan June 3 where a rest of another day was taken.

Detroit-Grand Rapids Air Mail Contract

By direction of the Postmaster General an advertisement has been issued inviting proposals for the operation of a contract air mail route in Michigan.

DETROIT, MICH., to GRAND RAPIDS, MICH., and RETURN, with such intermediate stops as may be agreed upon later.

Service to be not less than six round-trips per week, and on such a schedule as may be found most advantageous to the Department and the contractor. The base 100 miles each way.

The route is open to bidders regardless of residence, and bids will be received at the Department in Washington until noon of July 25, 1928.

The schedule to be adopted will require an average flying speed of approximately 80 m.p.h. The Department realizes that in some instances, due to weather conditions, etc., it may be impracticable to maintain such an average, but under favorable conditions even better time may be possible. Proper allowance will be made in all such cases.

The Department reserves the right to modify or change the schedule adopted if the needs of the service demand and operating conditions will permit.

Contracts are limited by law to not exceeding a four-year term.

Proposals should show the rate of compensation stated in percentage of revenue, which must not exceed 10%—the maximum allowed by law.

The postage rate over contract air mail routes is 10 cents an ounce or fraction thereof when the length of the route is not over 3,000 miles, 15 cents up to 50 and including 3,000 miles, 20 cents where the length is over 3,000 miles, with 5 cents additional for each mile traveled over the maximum—usual government operating rules.

Airplane Explores Dutch New Guinea

Prof. Mathew Bunting, head of the expedition which is exploring the interior of Dutch New Guinea, has reported by wireless, from the expedition's power camp at the foot of Mount Mendenhall, that a recent flight made with the Yushu Transport plane has been a complete success. This is the first flight of an airplane in this section of the coast, where civil rights of anyone almost unknown is confined.

The flight lasted an hour and a half and was productive of important geographical discoveries. An excellent reconnaissance of the territory was reported, as well as the capture of native villages. Troops, furnished by the Dutch East Indies Government, are taking the expedition.



The service airplane, Dutch East



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